

ABSTRACT

The invention relates to a turbine blade comprising a vane that runs along a blade axis and a platform region, which is located at the root of the vane and has a platform that extends transversally to the blade axis. The aim of the invention is to guarantee the improved cooling of a platform region and the transition region from a turbine blade to a platform of a turbine blade, thus ensuring the cooling of the delimitation of a flow channel in a gas turbine. To achieve this, the platform comprises a first platform wall that does not support the vane and a second platform wall that supports said vane. According to the inventive concept, at the root of the vane and over the course of the transition region from the turbine blade to the platform, said first platform wall is aerodynamically curved and the course of the second platform wall has a receding shoulder in relation to the first platform wall, as a continuation of the vane.